

# AUTOMATIC FLIGHT CONTROL SYSTEMS

## 3. ANALYSIS OF FEEDBACK CONTROL SYSTEMS

TEST INPUT SIGNAL:

Test input signal is technique of testing the stability of control systems by following given equations. There are four types of test function for testing the input signal.

☺STEP FUNCTION:

$$a(t) = 0 \quad \text{when } (t \leq 0)$$

$$a(t) = 1 \quad \text{when } (0 \leq t)$$

taking laplace of a f(t) signal  $\int_0^\infty f(t) e^{-st} dt$

$$L f(t) = \int_0^\infty f(t) e^{-st} dt = \int_0^\infty e^{-st} dt$$

$$R(s) = \frac{1}{s}$$

☺RAMP FUNCTION:

$$r(t) = 0 \quad \text{when } t < 0$$

$$r(t) = kt \quad \text{when } t > 0$$

taking laplace

$$L r(t) = \int_0^\infty kt e^{-st} dt = \frac{k}{s^2}$$

$$R(s) = \frac{1}{s^2} \quad (\text{for unit ramp } k=1)$$

$$R(s) = \frac{1}{s^2}$$

☺ PARABOLIC FUNCTION:

$$r(t) = 0 \quad \text{when } t < 0$$

$$r(t) = \frac{t^2}{2} \quad \text{when } t > 0$$

for unit parabolic  $k=1$

$$r(t) = \frac{t^2}{2} \quad t > 0$$

Laplace of function

$$L r(t) = \int_0^{\infty} t^2 \cdot e^{-st} dt = \frac{2}{s^3}$$

$$R(s) = \frac{2}{s^3}$$

☺ IMPULSE FUNCTION:

$$f(t) = 0 \quad \text{when } t \neq 0$$

$$f(t) = \infty \quad \text{when } t = 0$$

$$\int_{-\infty}^{\infty} f(t) dt = 1 \quad \text{after integration}$$

$$f(t) = \delta(t)$$

The pulse for which the duration tends to zero and amplitude tend to infinity is called impulse function.

TIME DOMAIN PERFORMANCE OF CHARACTERISTICS OF FEEDBACK CONTROL SYSTEMS:

$$\text{DAMPING RATIO: } \frac{\text{Real part of } s}{\text{Magnitude of } s} = \frac{-\zeta \omega_n}{\omega_n} = -\zeta$$

Underdamped case ( $0 < \zeta < 1$ )

Undamped case ( $\zeta = 0$ )

Critically damped case ( $\zeta = 1$ )

### EFFECT OF DERIVATIVE AND INTEGRAL CONTROL:

Derivative control action:

$$\frac{? (?)}{? (?)} = S k_d$$

$k_d$  is the derivative constant.  $E(s)$  is the input signal and  $M(s)$  is the output signal.

Integral control action:

$$\frac{? (?)}{? (?)} = \frac{? ?}{?}$$

$k_i$  is the integral constant.

Derivative and integral control functions increases the sensitivity of control systems.

### STEADY STATE ERROR:

☺ Static position error coefficient:

$$?_{??} = \frac{1}{1+??} \quad \text{where } k_p \text{ is the static position coefficient constant}$$

☺ Static velocity error coefficient:

$$?_{??} = \frac{1}{??} \quad \text{where } k_v \text{ is the velocity error coefficient constant}$$

☺ Static acceleration error coefficient:

$$?_{??} = \frac{?}{??} \quad \text{where } k_a \text{ is the acceleration error coefficient constant}$$

By Anand kumar Jha, student of aeronautical engineering